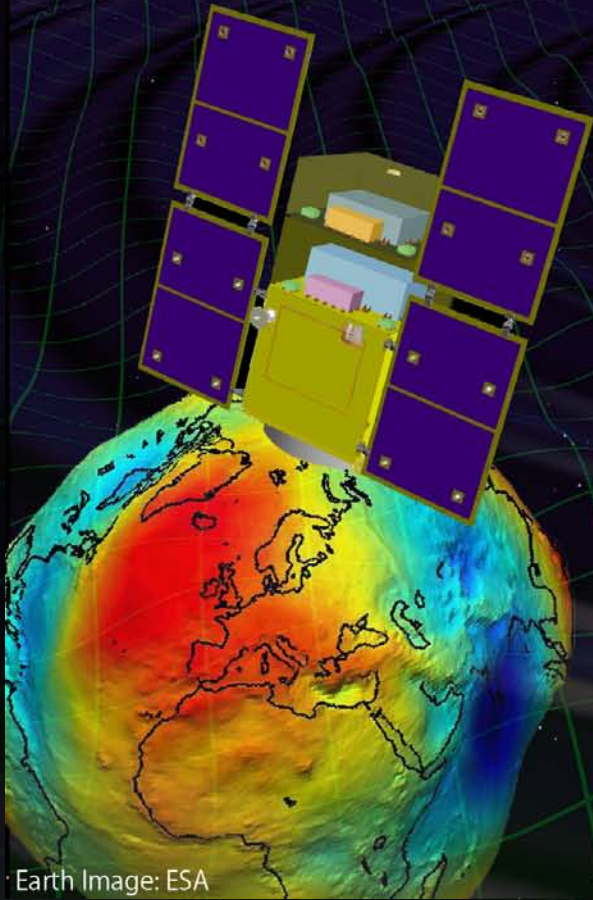


# Space Gravitational-Wave Antenna DECIGO



Earth Image: ESA



Original  
Picture : Sora

**Masaki Ando**

(Department of Physics, Kyoto University)

**Wataru Kokuyama**

(Department of Physics, University of Tokyo)

On behalf of  
DECIGO working group

GWADW2010 (May 20, 2010, Kyoto, Japan)

# 1. DECIGO

Overview and Science

Pre-conceptual Design

# 2. DECIGO Pathfinder

Overview and Science

Design and Status

Space Demonstration

← Wataru

← Kokuyama

# 3. Summary



# **1. DECIGO**

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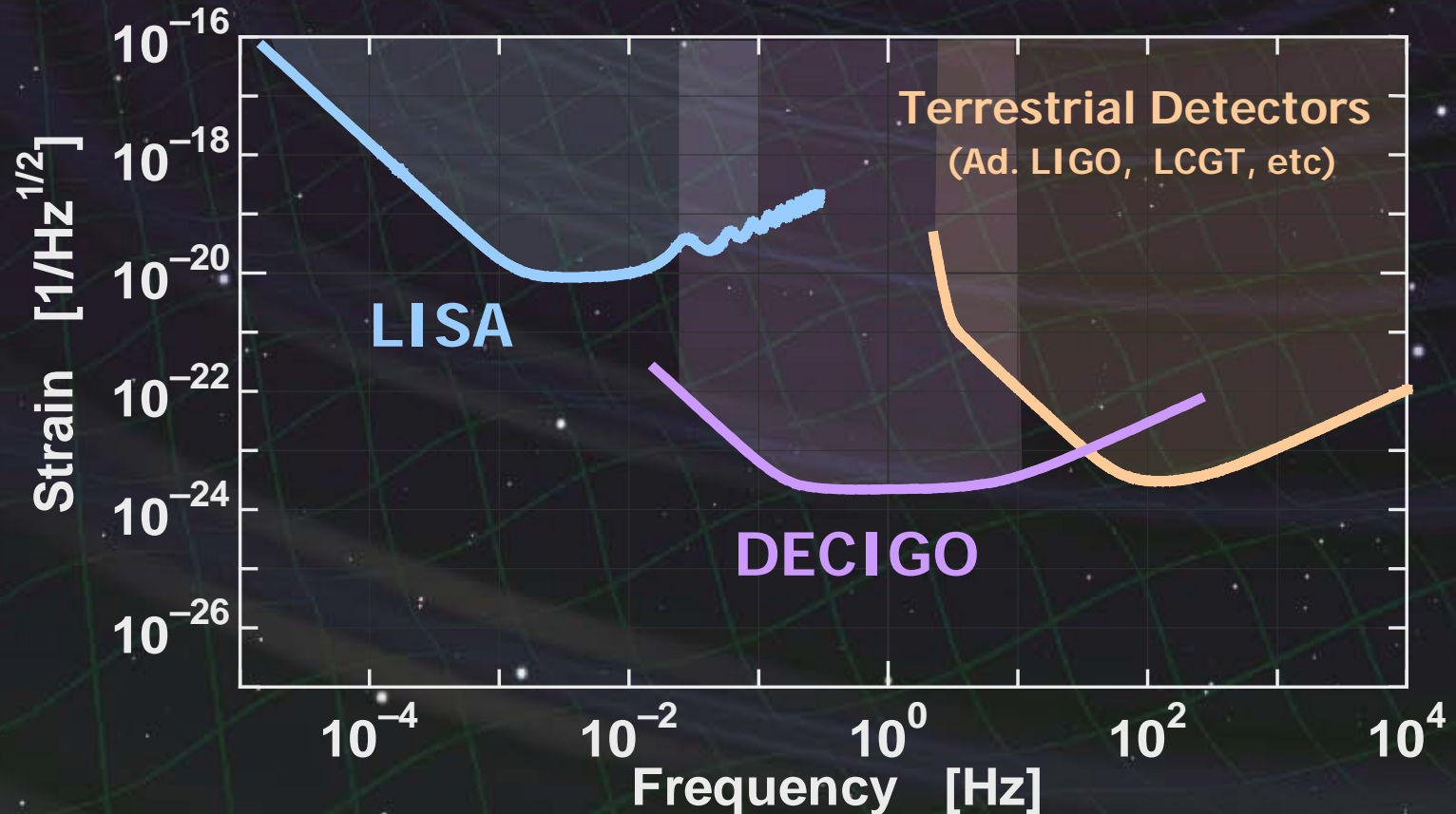
# **3. Summary**

**DECIGO** (Deci-hertz interferometer Gravitational wave Observatory)

Space GW antenna (~2027)  
Obs. band around 0.1 Hz



'Bridge' the obs. gap between  
**LISA** and **Terrestrial detectors**

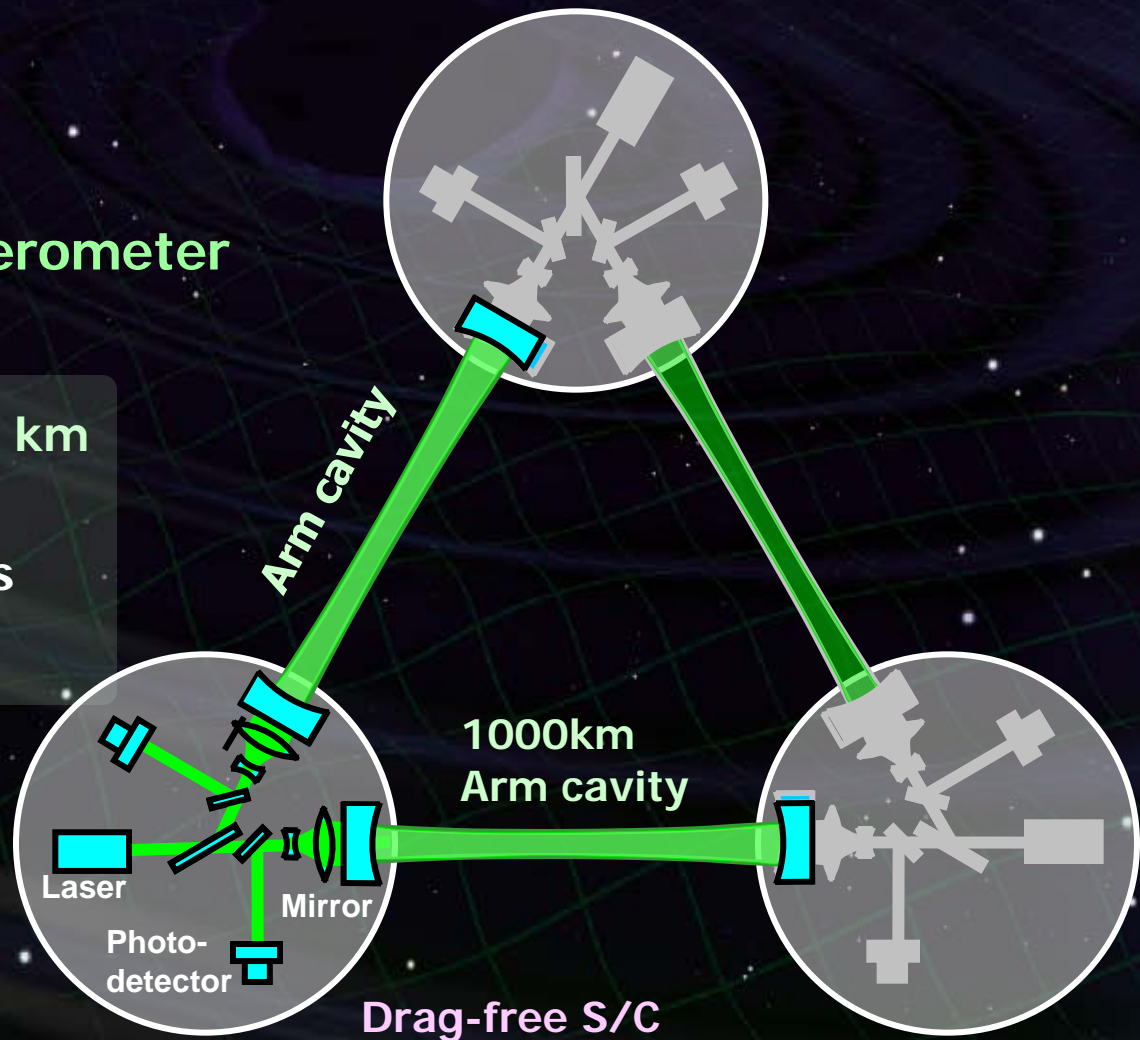


# DECIGO Interferometer



Interferometer Unit:  
Differential FP interferometer

Baseline length: 1000 km  
3 S/C formation flight  
3 FP interferometers  
Drag-free control

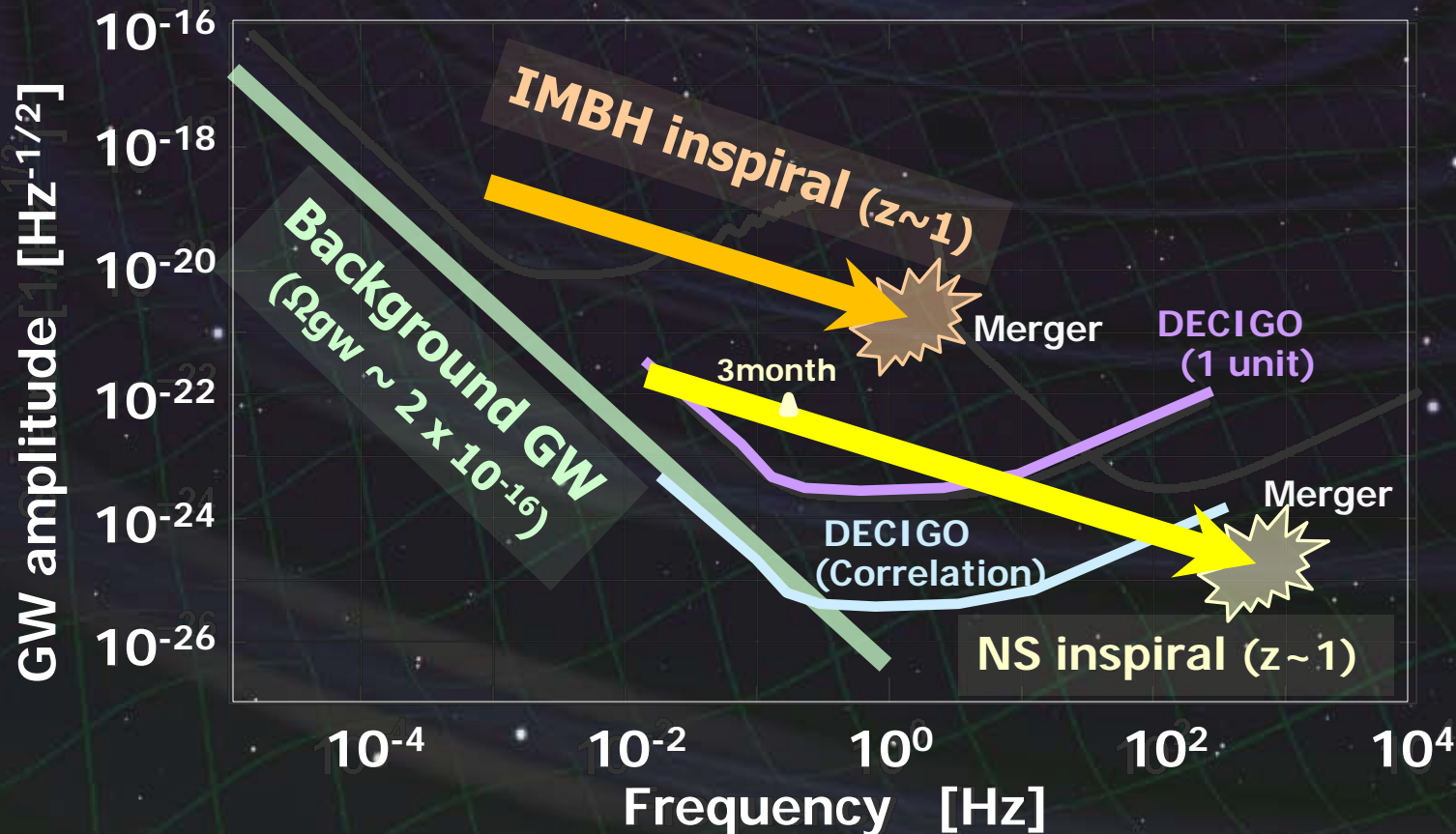


# Targets and Science

**IMBH** binary inspiral  
**NS** binary inspiral  
**Stochastic background**



Galaxy formation (Massive BH)  
Cosmology  
(Inflation, Dark energy)



# Constraint on dark energy

DECIGO will observe

$10^{4-5}$  NS binaries at  $z \sim 1$

↳ Precise 'clock' at cosmological distance

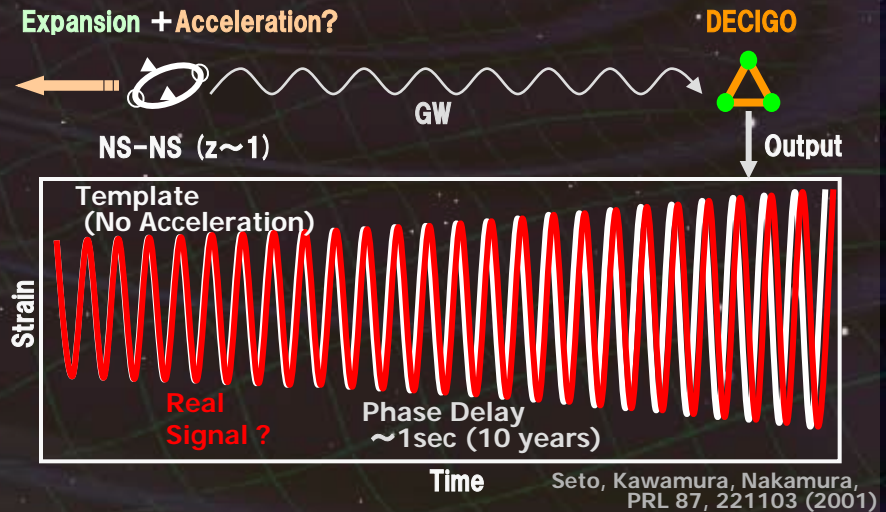
## 'Standard Siren'

Relationship between distance and redshift

Distance: chirp waveform

Redshift: host galaxy

→ Information on **acceleration of expansion of the universe**



Determine cosmological parameters

$$\Delta\Omega_m, \Delta\Omega_w, \Delta w \approx 1\%$$

**Absolute and independent measurement**

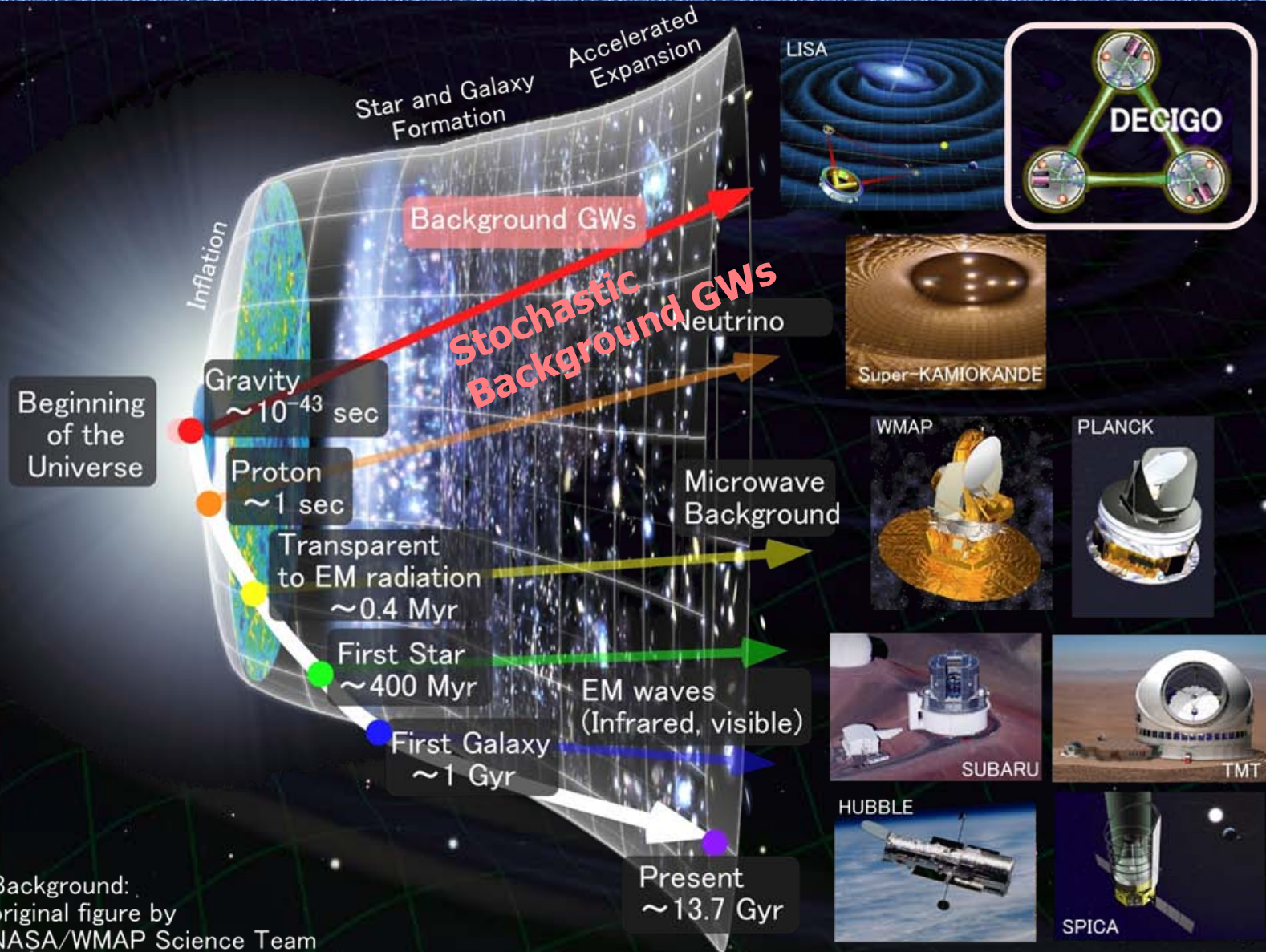
Angular resolution

$\sim 10$ arcmin (1 detector)

$\sim 10$ arcsec (3 detectors)

at  $z=1$

# Stochastic Background GWs



Background:  
original figure by  
NASA/WMAP Science Team



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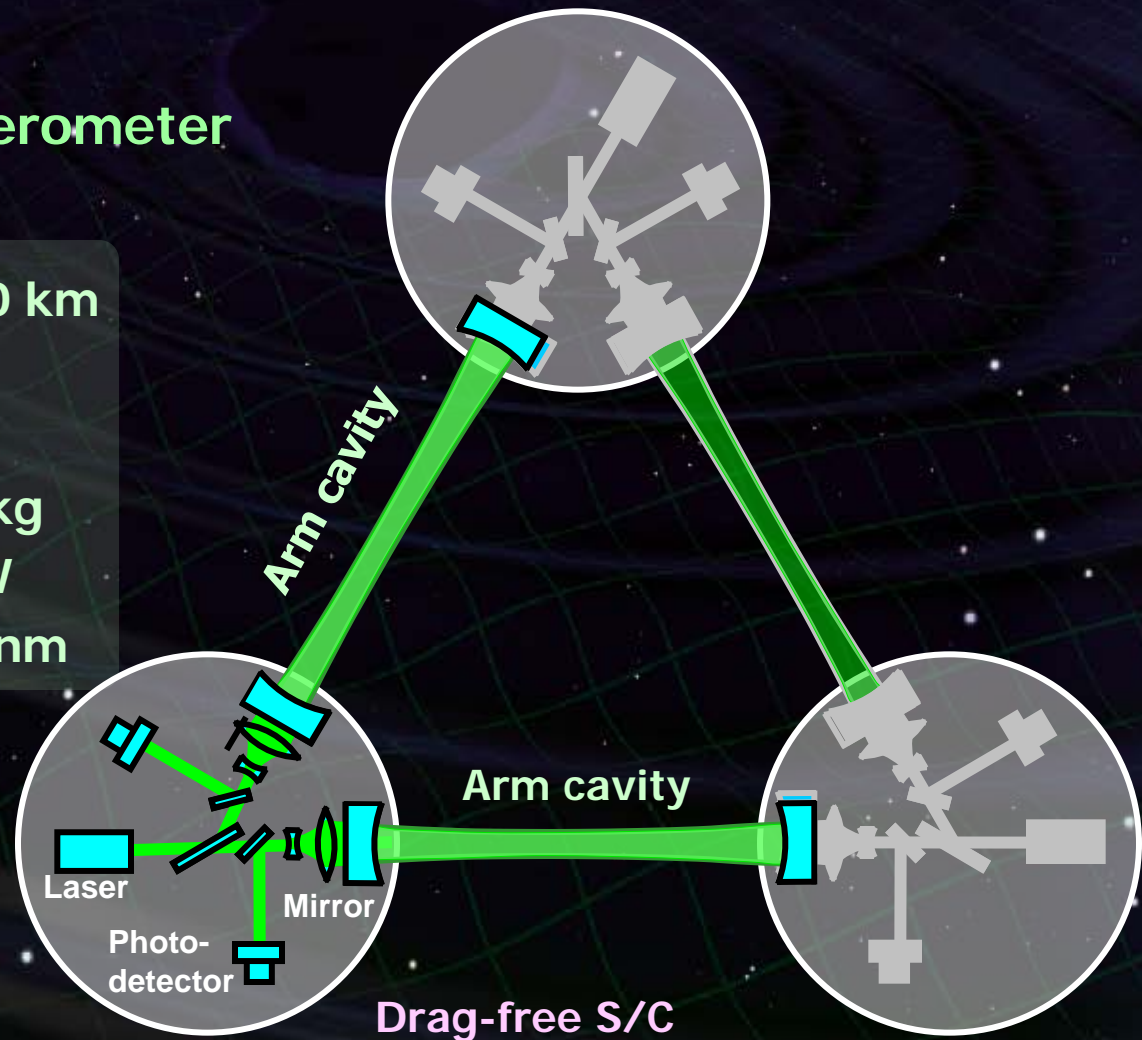
# 3. Summary

## Interferometer Unit:

### Differential FP interferometer

Arm length:	1000 km
Finesse:	10
Mirror diameter:	1 m
Mirror mass:	100 kg
Laser power:	10 W
Laser wavelength:	532 nm

S/C: drag free  
3 interferometers

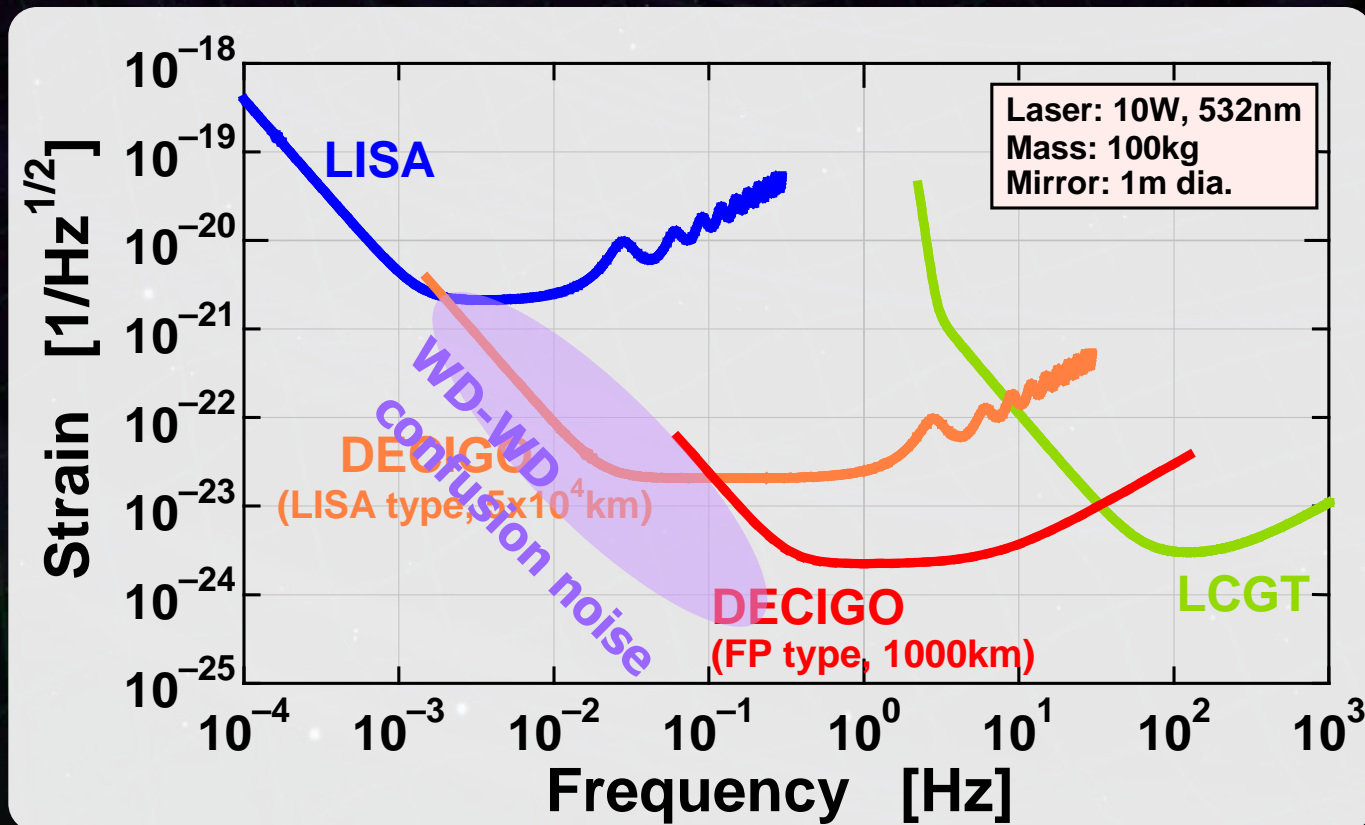


# Interferometer Design

## Transponder type vs Direct-reflection type

Compare : Sensitivity curves and Expected Sciences

⇒ Decisive factor: Binary confusion noise



# Cavity and S/C control

## Cavity length change

PDH error signal  $\rightarrow$  Mirror position (and Laser frequency)

Relative motion between mirror and S/C

Local sensor  $\rightarrow$  S/C thruster

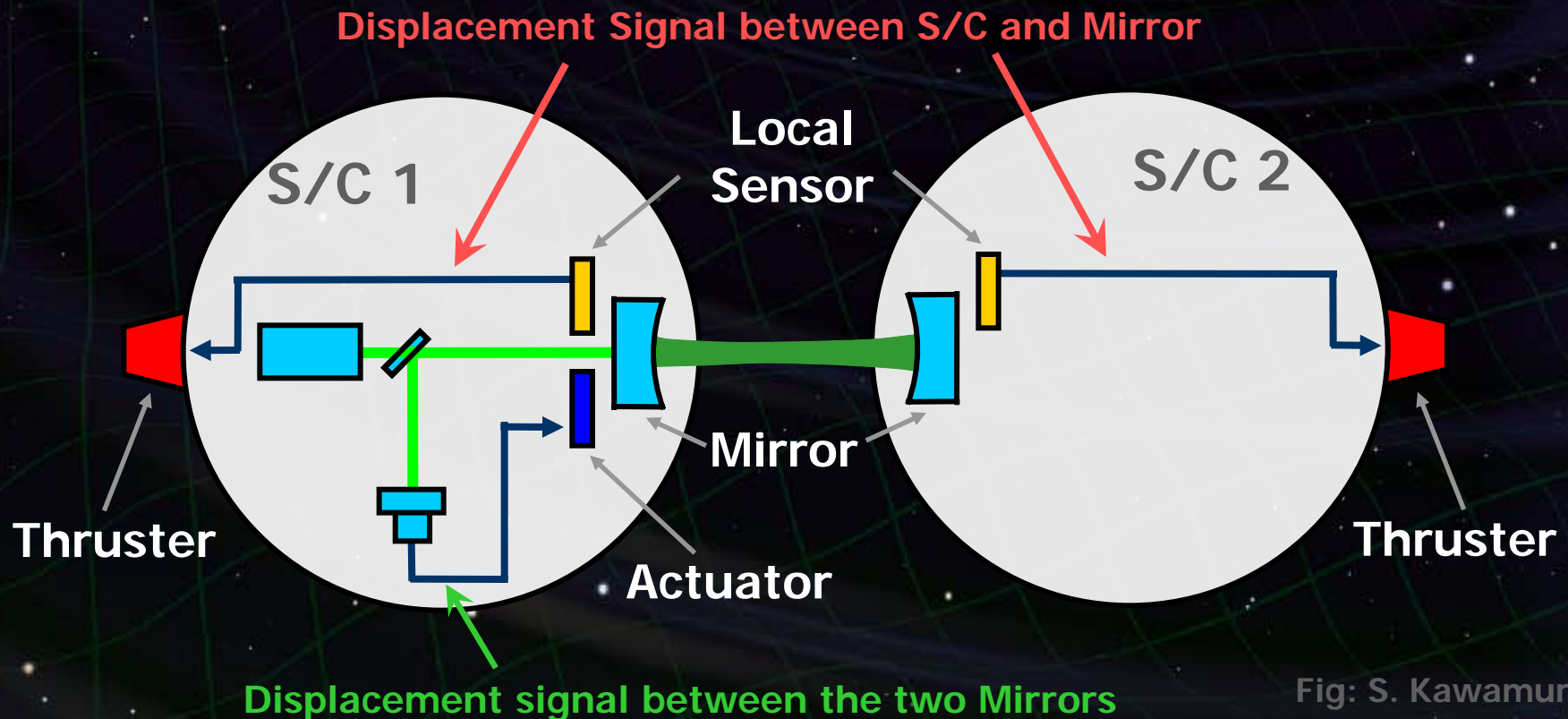


Fig: S. Kawamura

# Requirements

## Sensor Noise

Shot noise  $3 \times 10^{-18} \text{ m/Hz}^{1/2}$  (0.1 Hz)

⇒ x 10 of LCGT in phase noise

Other noises should be well below the shot noise

Laser freq. noise:  $1 \text{ Hz/Hz}^{1/2}$  (1Hz)

Stab. Gain  $10^5$ , CMRR  $10^5$

## Acceleration Noise

Force noise  $4 \times 10^{-17} \text{ N/Hz}^{1/2}$  (0.1 Hz)

⇒ x 1/50 of LISA

External force sources

Fluctuation of magnetic field, electric field,  
gravitational field, temperature, pressure, etc.

# Orbit and Constellation

## Candidate of orbit:

Record-disk orbit around the Sun

Relative acc.  $4 \times 10^{-12} \text{ m/s}^2$   
(Mirror force  $\sim 10^{-9} \text{ N}$ )

Halo orbit around L2 (or L1)

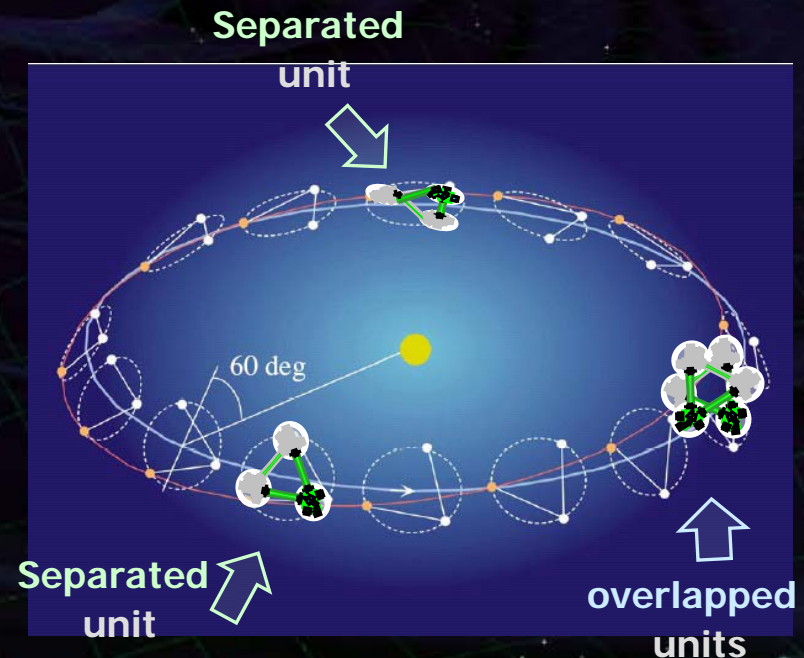
Relative acc.  $4 \times 10^{-7} \text{ m/s}^2$   
(Mirror force  $\sim 10^{-4} \text{ N}$ )

## Constellation

4 interferometer units

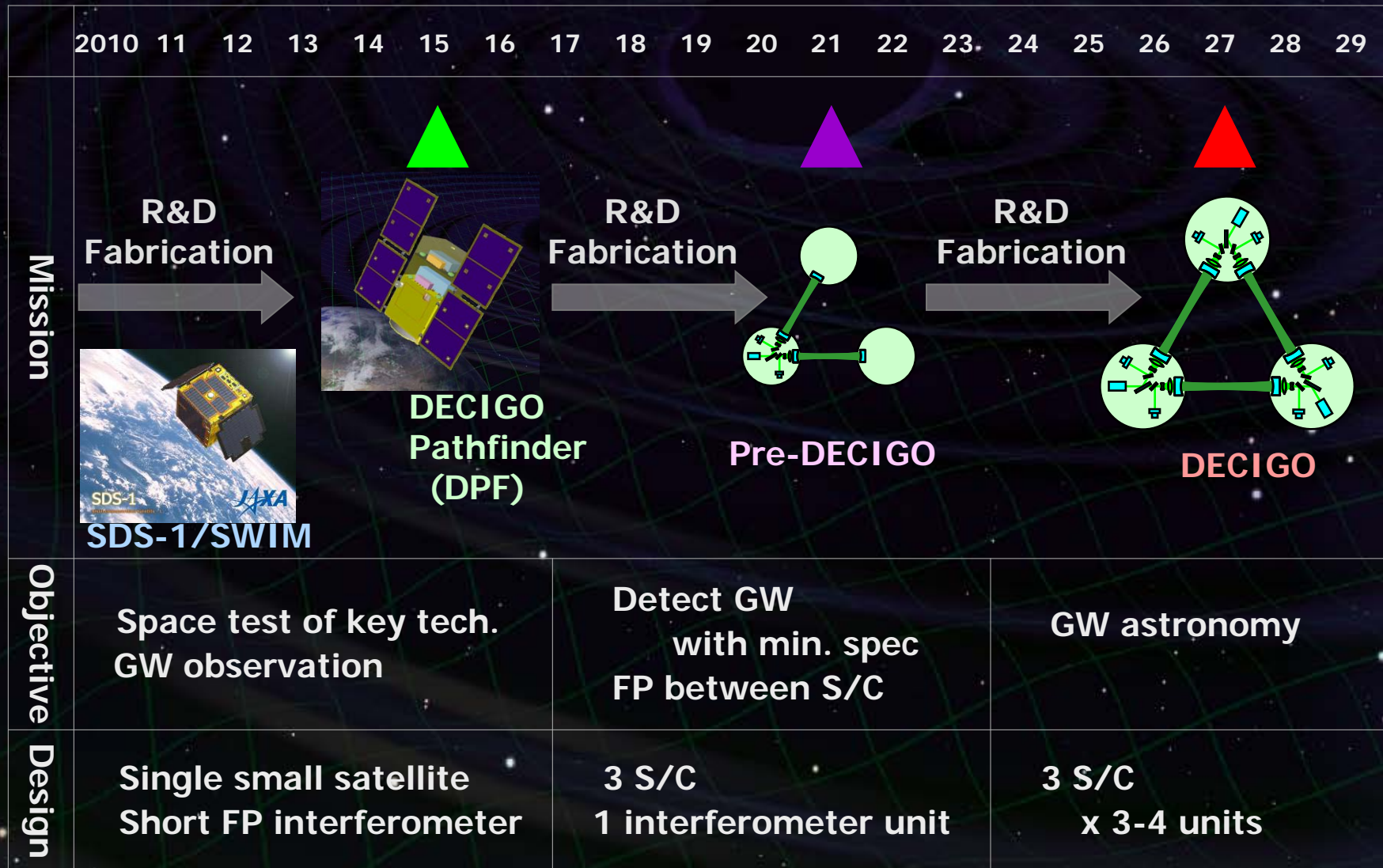
2 overlapped units  $\rightarrow$  Cross correlation

2 separated units  $\rightarrow$  Angular resolution



# Roadmap

Figure: S.Kawamura



# Organization



**PI: Kawamura (NAOJ)**  
**Deputy: Ando (Kyoto)**

**Executive Committee**  
Kawamura (NAOJ), Ando (Kyoto), Seto (Kyoto), Nakamura (Kyoto), Tsubono (Tokyo), Tanaka (Kyoto), Funaki (ISAS), Numata (Maryland), Sato (Hosei), Kanda (Osaka city), Takashima (ISAS), Ioka (KEK), Yokoyama (Tokyo)

**Pre-DECIGO**  
Sato (Hosei)

**Detector**  
Akutsu (NAOJ)  
Numata (Maryland)

**Science, Data**  
Tanaka (Kyoto)  
Seto (Kyoto)  
Kanda (Osaka city)

**Satellite**  
Funaki (ISAS)

## Design phase

**DECIGO pathfinder**  
**Leader: Ando (Kyoto)**

## Mission phase

**Detector**  
Sato (Hosei)  
Ueda (NAOJ)  
Aso (Tokyo)

**Laser**  
Musha (ILS)  
Ueda (ILS)

**Drag free**  
Moriwaki (Tokyo)  
Sakai (ISAS)

**Thruster**  
Funaki (ISAS)

**Bus**  
Takashima (ISAS)

**Data**  
Kanda (Osaka city)



- Supports from **LISA**  
Technical advices from LISA/LPF experiences  
Support Letter for DECIGO/DPF, Joint workshop (2008.11)
- Collab. with **Stanford univ. group**  
Drag-free control of DECIGO/DPF  
UV LED Charge Management System for DPF
- Collab. with **NASA/GSFC**  
Fiber Laser , started discussion
- Collab. with **JAXA navigation-control section**  
→ formation flight of DECIGO, DPF drag-free control
- Research Center for the Early Universe (**RESCEU**), Univ. of Tokyo  
Support DECIGO as ones of main projects (2009.4-)
- Advanced technology center ( **ATC**) of **NAOJ**  
Will make it a main nucleus of DPF

# 1. DECIGO

Overview and Science  
Pre-conceptual Design



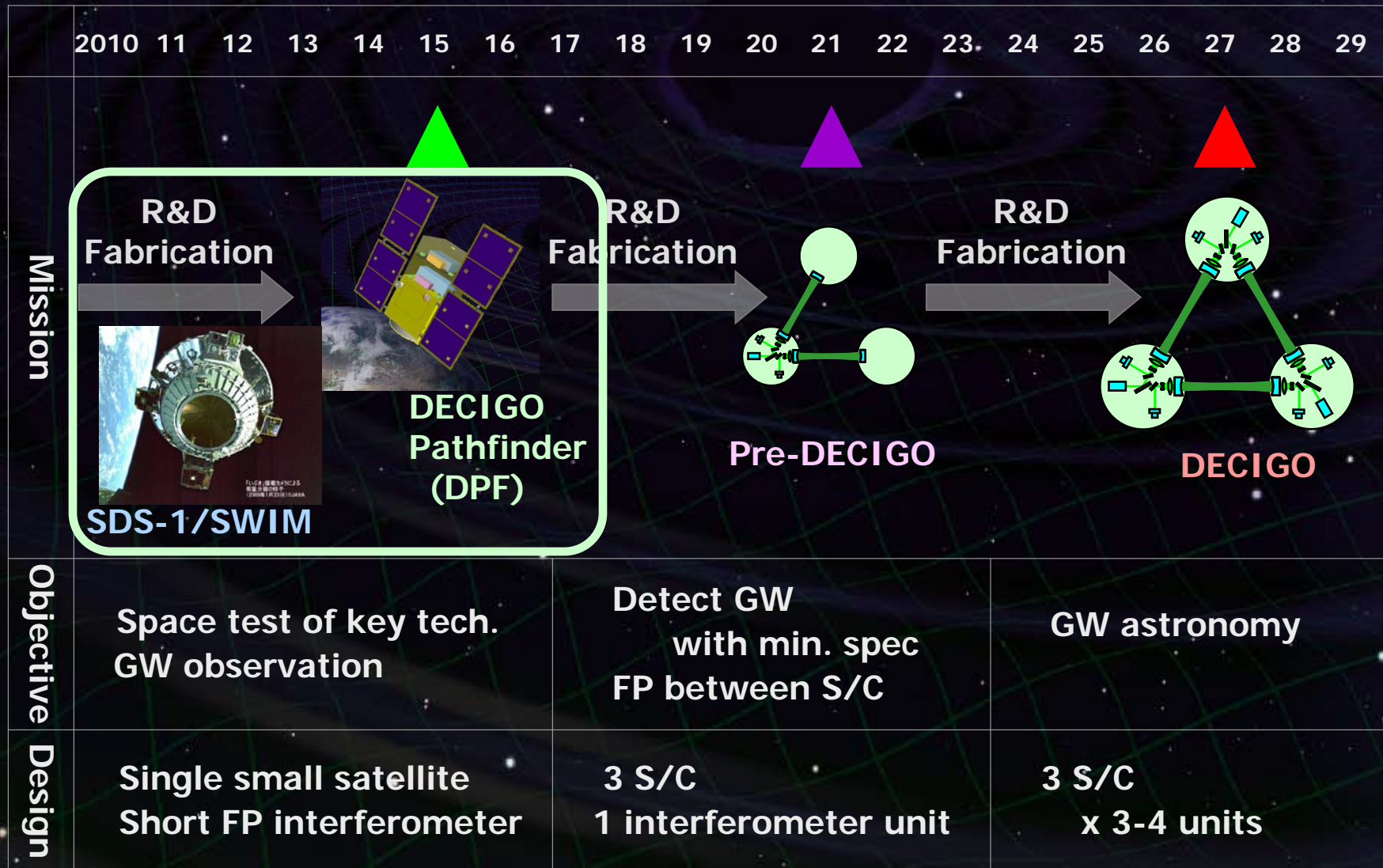
# 2. DECIGO Pathfinder

Overview and Science  
Design and Status  
Space Demonstration

# 3. Summary

# Roadmap

Figure: S.Kawamura



## DECIGO Pathfinder (DPF)

First milestone mission for DECIGO

Shrink arm cavity

DECIGO 1000km → DPF 30cm

### Single satellite

(Payload ~ 1m<sup>3</sup> , 350kg)

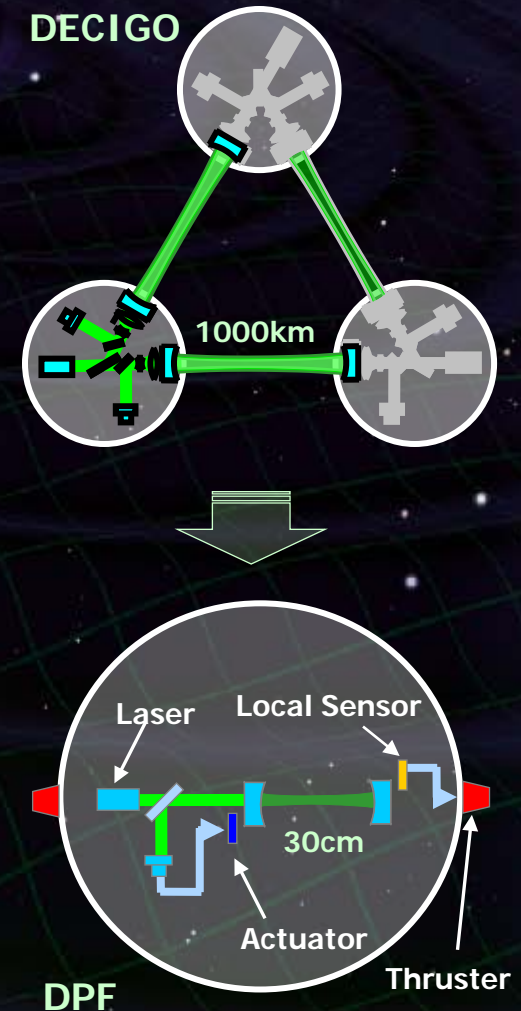
### Low-earth orbit

(Altitude 500km, sun synchronous)

30cm FP cavity with 2 test masses

Stabilized laser source

Drag-free control



# DPF satellite

## DPF Payload

Size : 950mm cube  
Weight : 150kg  
Power : 130W  
Data Rate: 800kbps  
Mission thruster x12

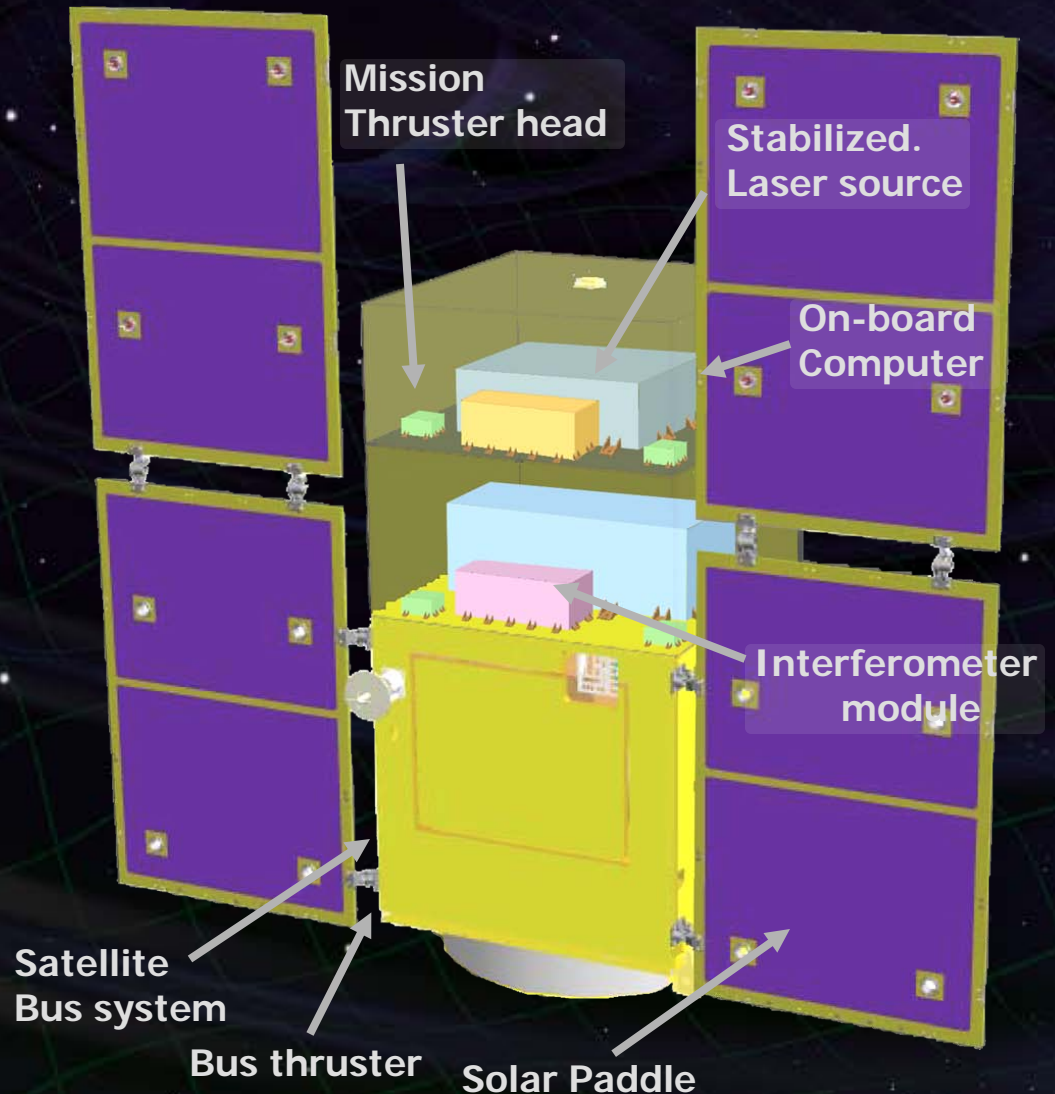
Power Supply  
SpW Comm.



## Satellite Bus

('Standard bus' system)

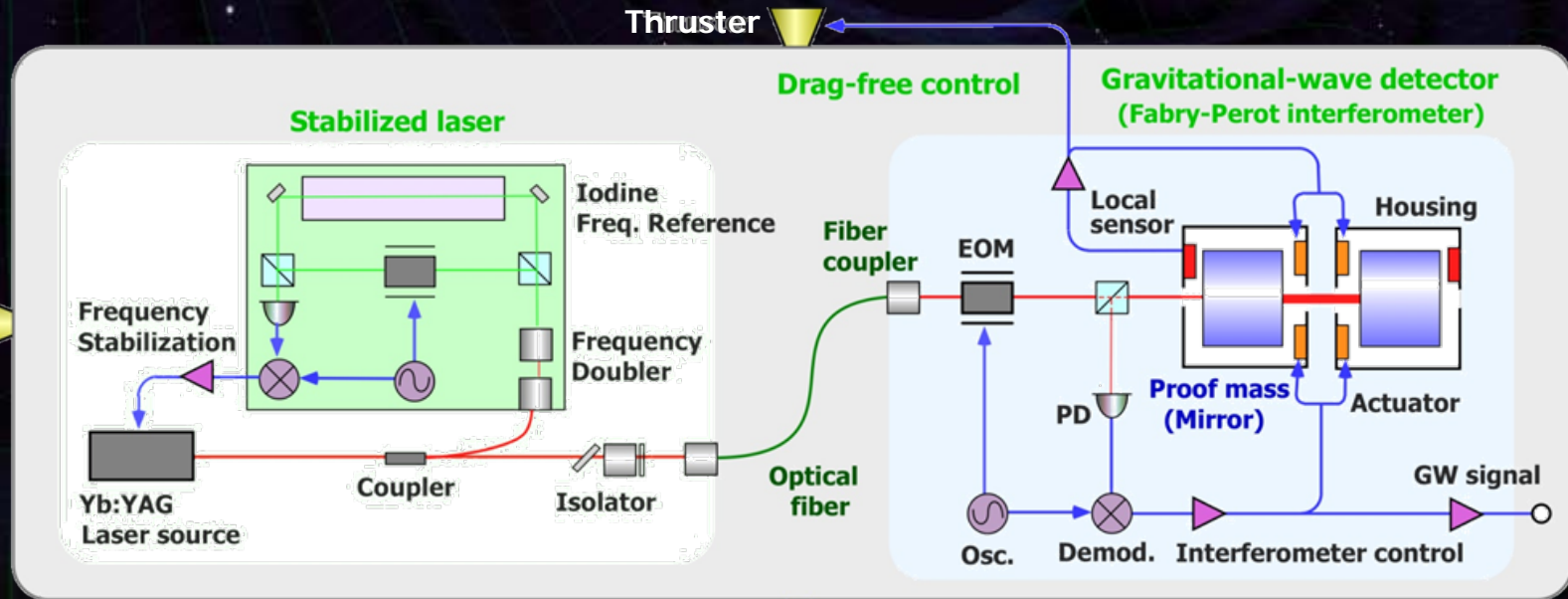
Size :  
950x950x1100mm  
Weight : 200kg  
SAP : 960W  
Battery: 50AH  
Downlink : 2Mbps  
DR: 1GByte  
3N Thrusters x 4



# DPF mission payload

Mission weight : ~ 150kg  
Mission space : ~ 95 x 95 x 90 cm

Drag-free control  
Local sensor signal  
→ Feedback to thrusters



## Laser source

Yb:YAG laser (1030nm)  
Power : 25mW  
Freq. stab. by Iodine abs. line

## Fabry-Perot interferometer

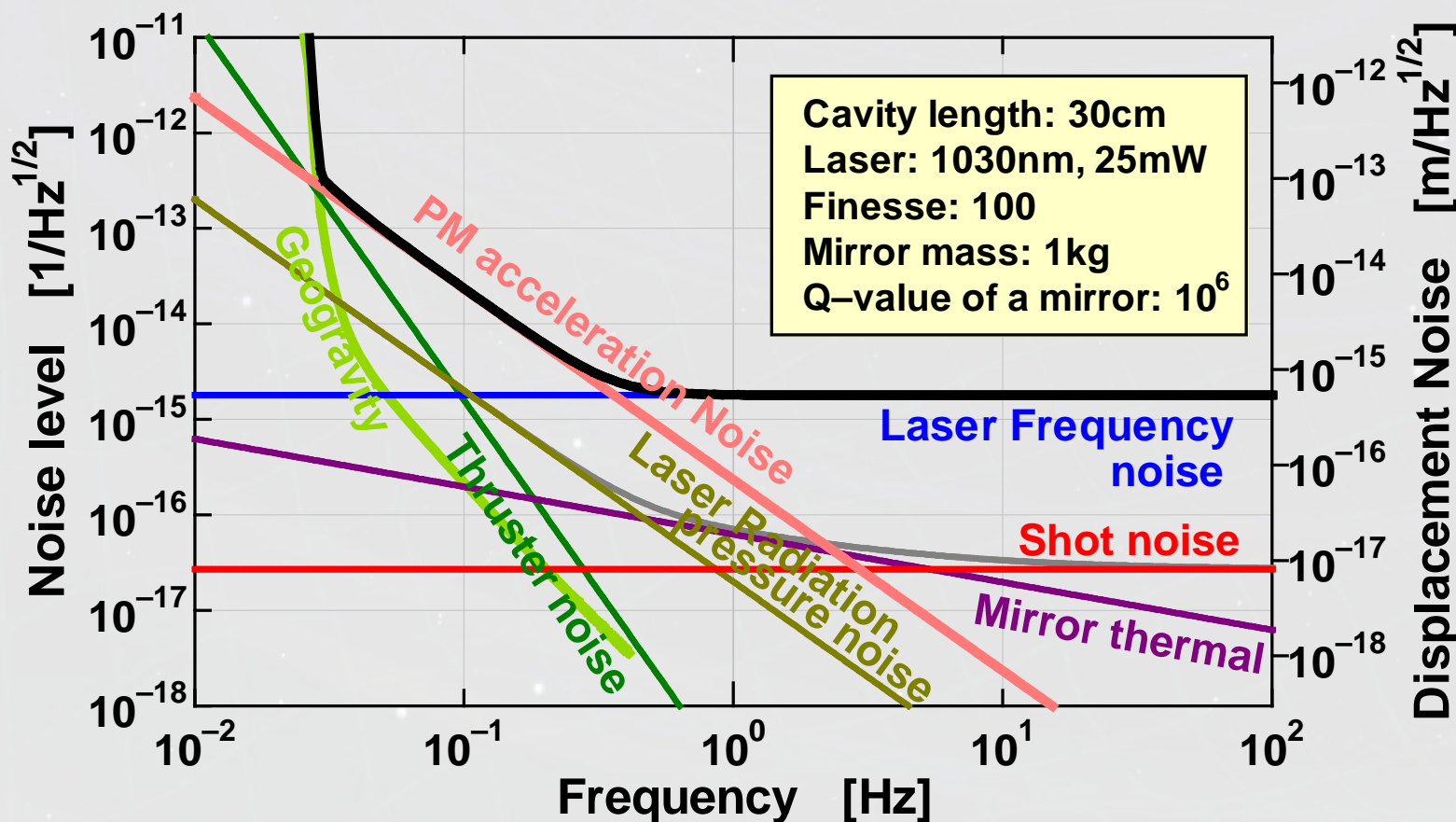
Finesse : 100  
Length : 30cm  
Test mass : ~ 1kg  
Signal extraction by PDH

# DPF Sensitivity

Laser source : 1030nm, 25mW  
IFO length : 30cm  
Finesse : 100, Mirror mass : 1kg  
Q-factor :  $10^5$ , Substrate: TBD  
Temperature : 293K

Satellite mass : 350kg, Area: 2m<sup>2</sup>  
Altitude: 500km  
Thruster noise:  $0.1\mu\text{N}/\text{Hz}^{1/2}$

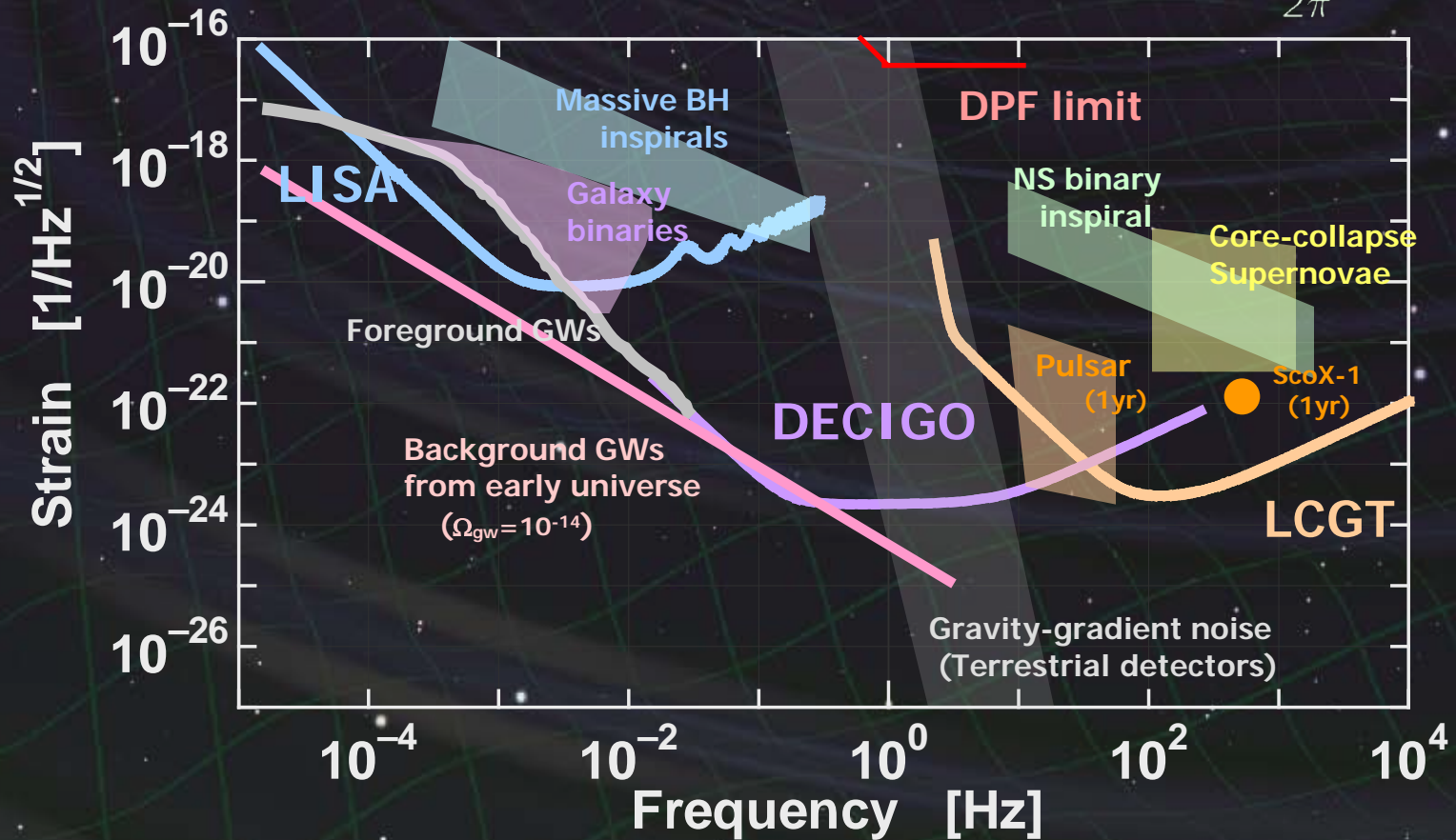
(Preliminary parameters)



# DPF sensitivity

DPF sensitivity  $h \sim 2 \times 10^{-15} \text{ Hz}^{1/2}$   
 (x10 of quantum noises)

$$f \sim \frac{1}{2\pi} \sqrt{GM/R^3}$$





# GW target of DPF

## Blackholes events in our galaxy

### IMBH inspiral and merger

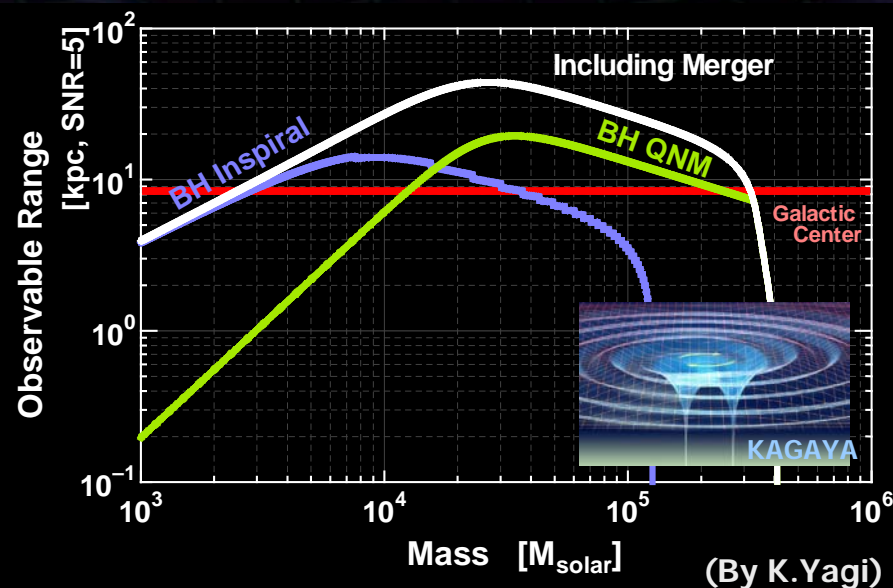
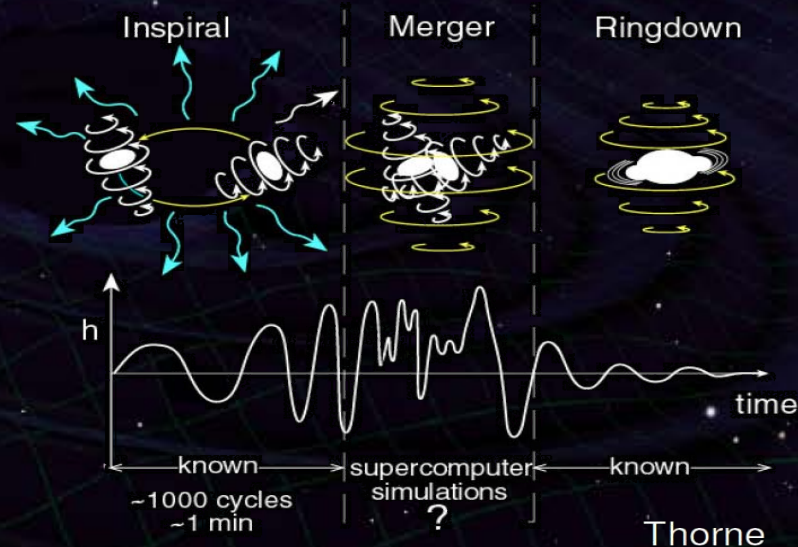
$h \sim 10^{-15}$ ,  $f \sim 4$  Hz  
Distance 10kpc,  $m = 10^3 M_{\text{sun}}$   
Obs. Duration ( $\sim 1000$ sec)

### BH QNM

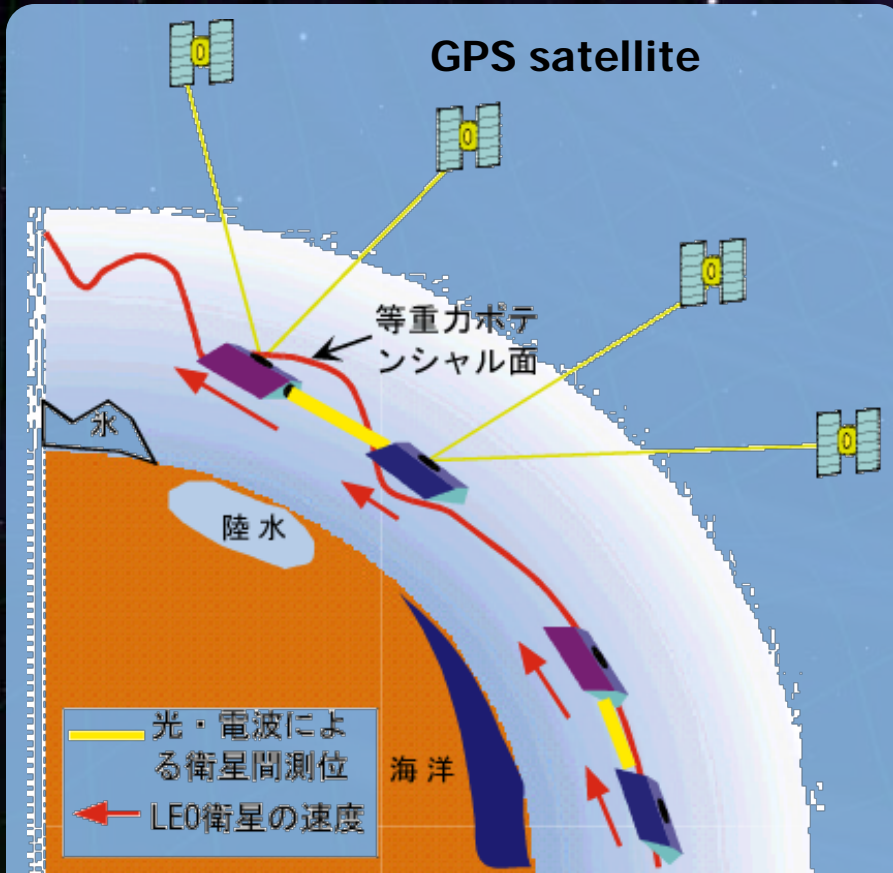
$h \sim 10^{-15}$ ,  $f \sim 0.3$  Hz  
Distance 1Mpc,  $m = 10^5 M_{\text{sun}}$

Observable range covers  
our Galaxy (SNR  $\sim 5$ )

Hard to access by others  
→ Original observation



## Measure gravity field of the Earth from Satellite Orbits, and gravity-gradiometer



By Araya and Fukuda

Determine global gravity field  
→ Density distribution  
Monitor of change in time  
Ground water motion  
Strains in crusts by  
earthquakes and volcanoes

Observation Gap  
between GRACE and GRACE-FO  
(2012-16)  
→ DPF contribution  
in international network

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Overview and Science  
**Design and Status**  
Space Demonstration

# 3. Summary



# Interferometer Module

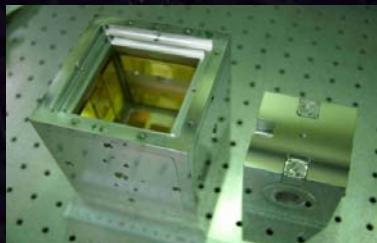
## Interferometer Module : Test mass + IFO

### Test-mass module

→ Gravity reference

- BBM of Module, Sensor, Actuator, Clump/Release
- $\mu$ -Grav. Exp.

Hosei, NAOJ, Ochanomizu, Stanford



### Interferometer

→ GW, Gravity observation

- 30cm IFO BBM
- Digital control
- Packaging
- Monolithic Opt.



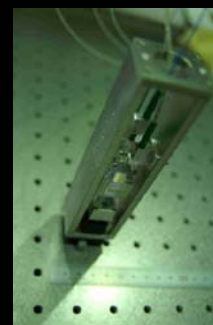
NAOJ, U-Tokyo

### Laser sensor

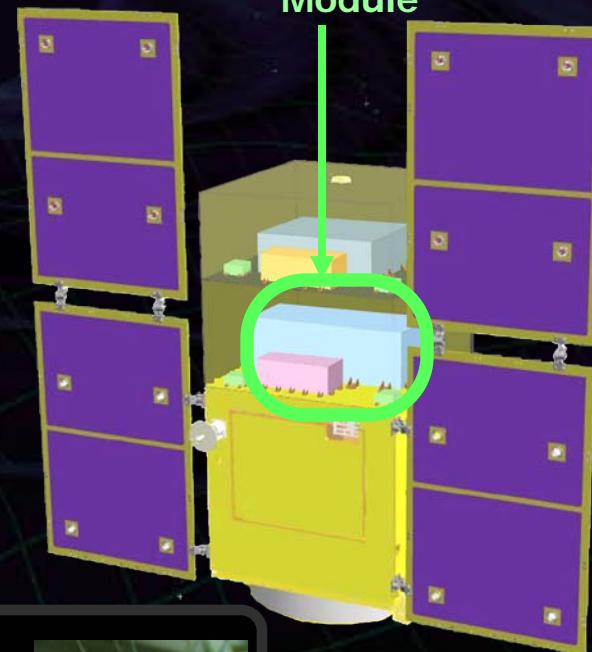
→ Small MI

- BBM test
- Sensitivity meas.

ERI, U-Tokyo



Interferometer Module



# Stabilized Laser Module

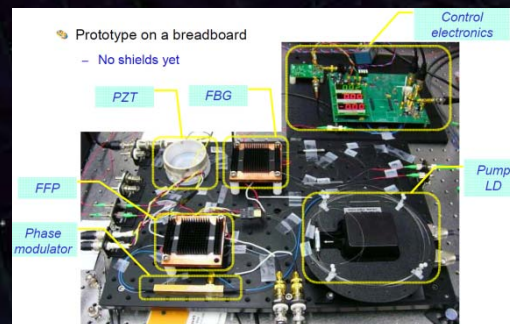
## Stabilized Laser : Laser source + Stabilization system

Yb:YAG (NPRO or Fiber laser)

→ Laser source

- BBM development

UEC, NASA/GSFC



I<sub>2</sub> absorption line  
→ Frequency reference

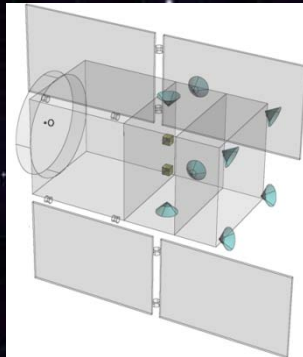
- BBM development
- Stability meas.

UEC, NICT



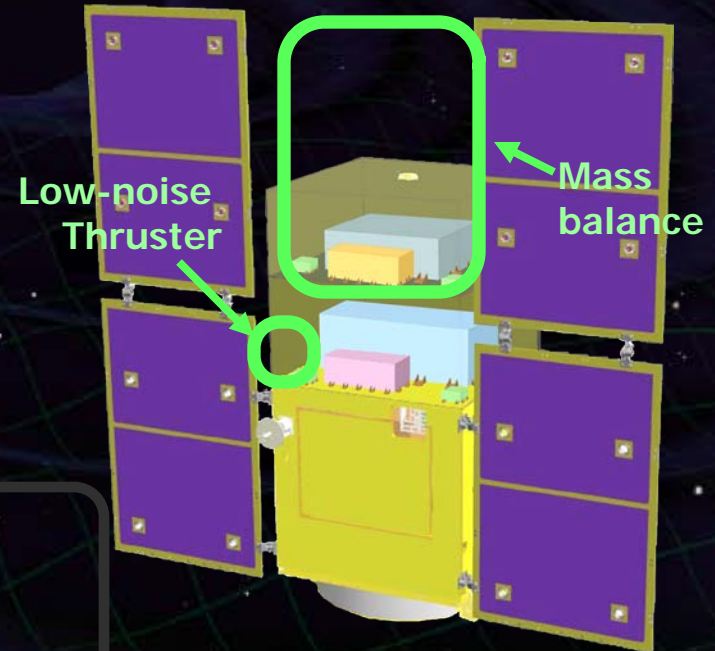
## Attitude and Drag-free control : Structure, Thrusters, Control

### Structure, thermal stability



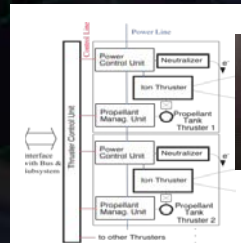
- Passive attitude stability
- Drag-free control

U-Tokyo, JAXA



### Low-noise Thruster

→ Actuators for satellite control



- BBM and system design

JAXA, NDAJ, Tokai-U

## Signal Processing and Control : SpaceWire-based system

SpC2 + SpW system

→ Signal processing and install. ctrl



Space demonstration  
by SDS-1/SWIM



JAXA, U-Tokyo, Kyoto

SWIMmn demonstration

→ Test mass control in orbit

**TAM: Torsion Antenna Module with free-falling test mass**  
(Size : 80mm cube, Weight : ~500g)

**Test mass**  
~47g Aluminum, Surface polished  
Small magnets for position control

**Coil**  
Used for test-mass position control  
Max current ~10mA

**Photo sensor**  
Reflective-type optical displacement sensor  
Separation to mass ~1mm  
Sensitivity ~ 10<sup>-6</sup> m/Hz<sup>1/2</sup>  
6 PSDs to monitor mass motion

z Error [mm]

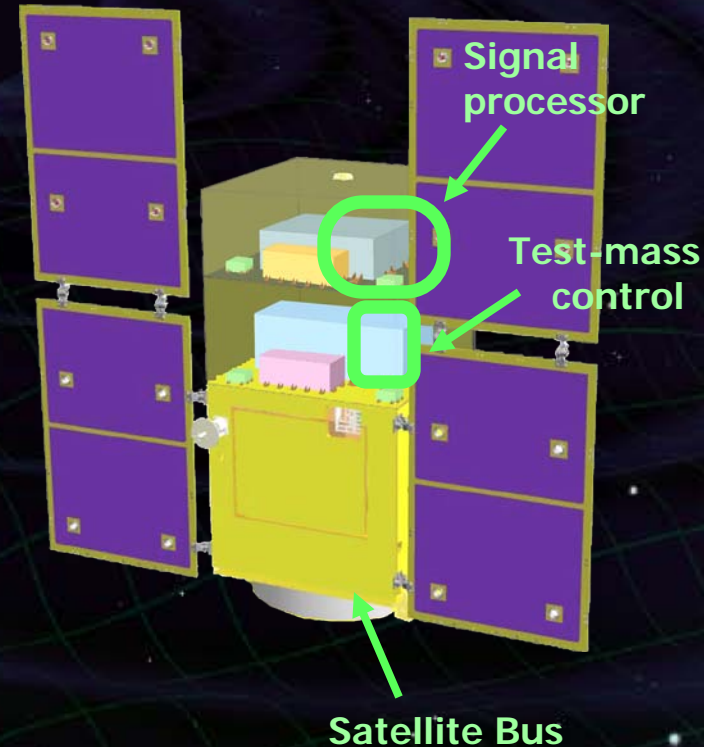
z Feedback [V]

Yaw Error [deg]

Yaw Feedback [V]

Time [sec]

JAXA, U-Tokyo, Kyoto



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⇒ **Space Demonstration**

# 3. Summary



# DPF mission status

DPF : One of the candidate of  
JAXA's small satellite series



At least 3 satellite in 5 years with  
Standard Bus + M-V follow-on rocket

1<sup>st</sup> mission (2012): SPRINT-A/EXCEED

2<sup>nd</sup> mission (~2013/14) : ERG  
DPF survived until final two

3<sup>rd</sup> mission (~2015/16) : TBD

DPF is one of the strongest  
candidates of the 3<sup>rd</sup> mission



SPRINT-A /EXCEED  
UV telescope mission



Next-generation  
Solid rocket booster (M-V FO)  
Fig. by JAXA

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# 3. Summary

## DECIGO : Fruitful Sciences

Very beginning of the Universe

Dark energy

Galaxy formation

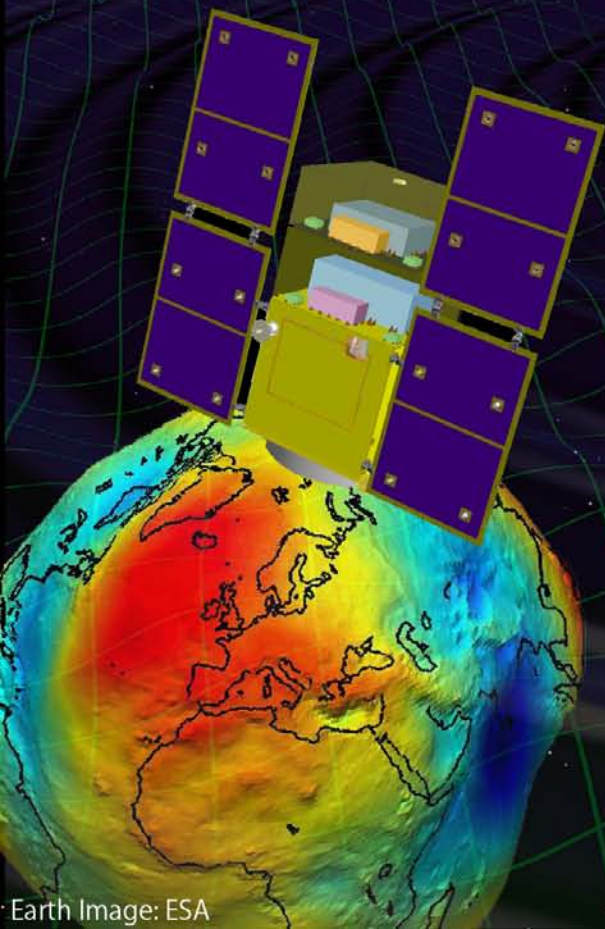
## DECIGO Pathfinder

Important milestone for DECIGO

Strong candidate of JAXA's satellite series

**SWIM** – under operation in orbit

**first precursor to space!**



Earth Image: ESA

End



Original  
Picture : Sora